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PROPULSION 3 cf. Eq. 2-7 of Biblarz and Sutton (2001) [1]. De ne some quantities: mass ratio = m_f / m_0 where $m_f :=$ nal mass (after rocket operation had consumed all usable propellant), which is M_0 above $m_0 :=$ mass before rocket operation, which is $M(0)$ propellant mass fraction $m_p / M(0)$ cf. Eq. 2-8 of Biblarz and Sutton (2001) [1]. For thrust F_{thrust} , $F_{thrust} = \dot{m} u_e$

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